

Abstract

The invention relates to a compressor, which is axially flowed through, for a gas turbine having an axially displaceable rotor. An annular flow channel, which narrows in an axial direction, is formed between a rotationally fixed outer delimiting surface and an inner delimiting surface on the rotor. A stationary ring comprised of guide profiles and at least one ring comprised of moving profiles attached to the rotor are placed inside said annular flow channel. The end of each moving or guide blade is located opposite an axial section of one of both delimiting surfaces while forming a radial gap. The aim of the invention is to provide a non-positive-displacement machine having an axially displaceable rotor whose velocity losses are at least not increased during an axial displacement of the rotor. To this end, the invention provides that the size of the radial gap between the end of each moving or guide blade and the opposite axial section of the delimiting surface is constant at least over the path of displacement of the rotor, and the radial gap extends parallel to the rotation axis of the rotor.